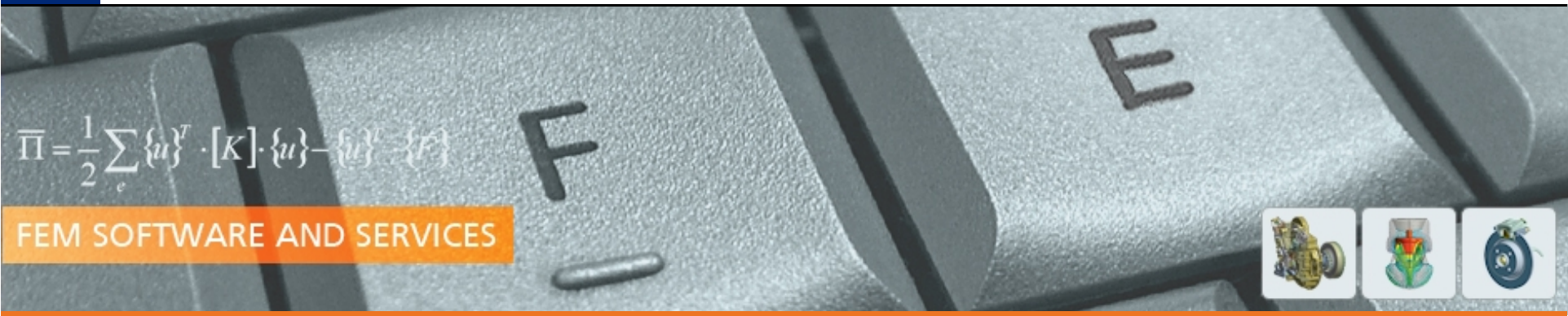

$$\bar{\Pi} = \frac{1}{2} \sum_e \{u\}^T \cdot [K] \cdot \{u\} - \{u\}^T \cdot \{F\}$$

FEM SOFTWARE AND SERVICES



# Efficient Design of Composite Rotor Blades using ANSYS

Dipl.-Ing. Steffen Schiele, CADFEM GmbH  
[sschiele@cadfem.de](mailto:sschiele@cadfem.de)


$$\bar{\Pi} = \frac{1}{2} \sum_e \{u\}^T \cdot [K] \cdot \{u\} - \{u\}^T \cdot \{F\}$$

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# Introduction to Lightweight Materials

Usage and Basics

# Introduction to Lightweight Materials/Structures



sports & recreation



automobile



Monkeylite Composite Riser Bar



aerospace



sailplane



wind energy

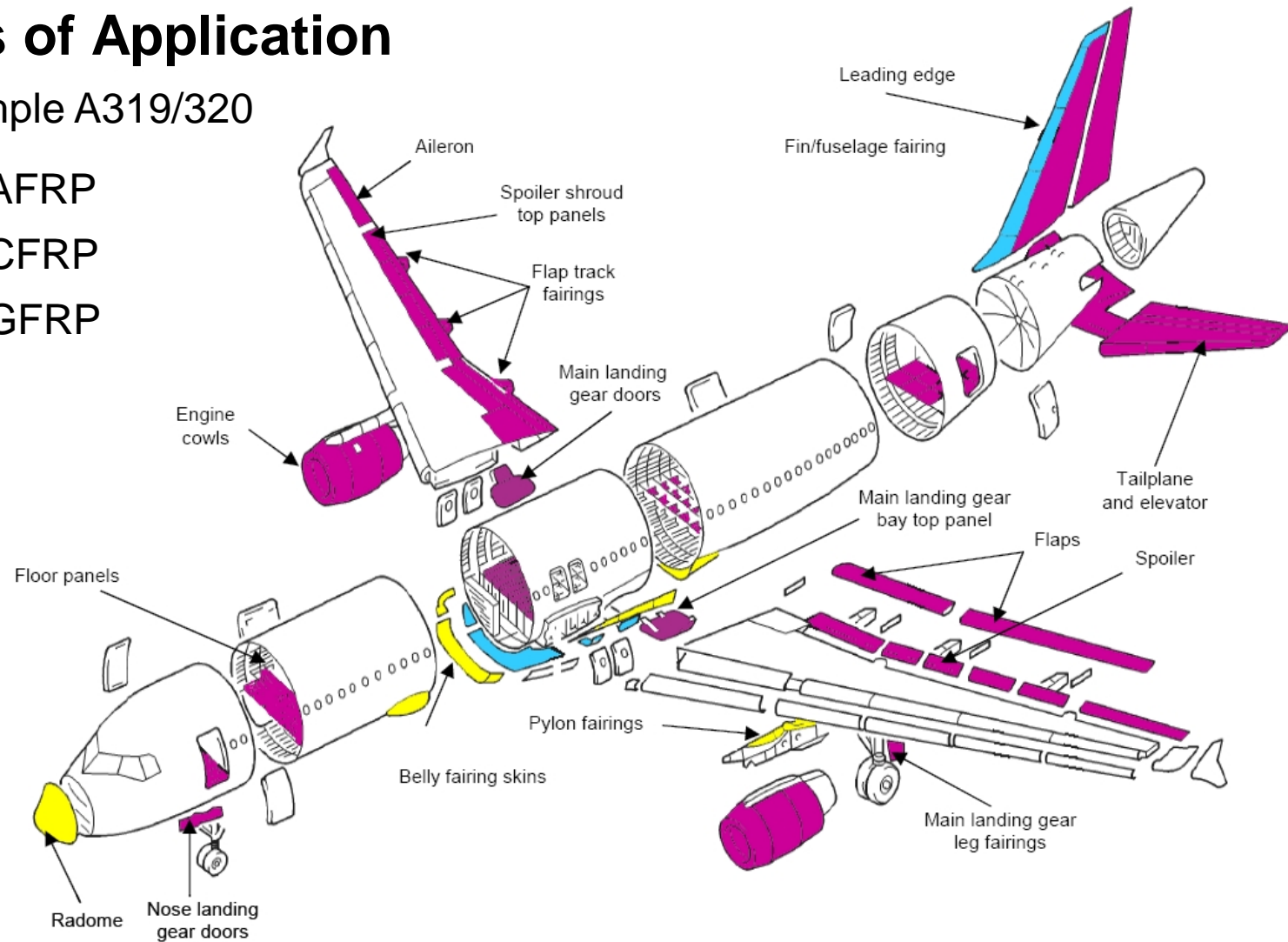


# Introduction to Lightweight Materials/Structures

## Areas of Application

Example A319/320

- AFRP
- CFRP
- GFRP





# Introduction to Lightweight Materials/Structures

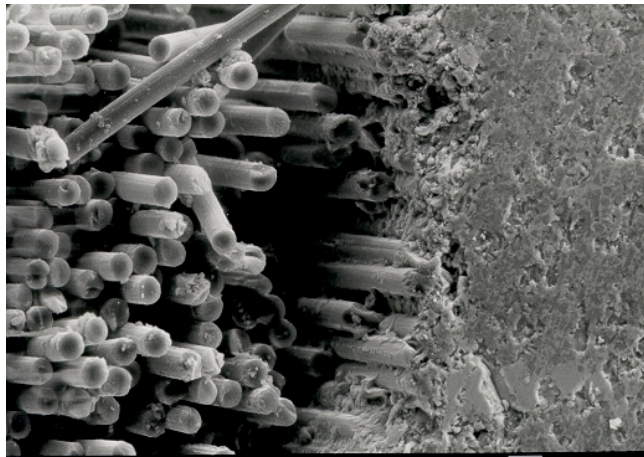
## Areas of Application



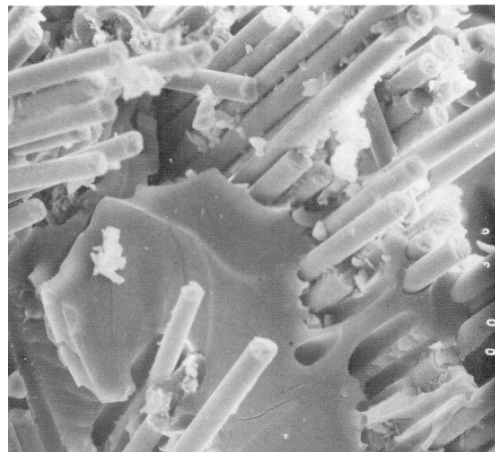
# Introduction to Composites

## Notion Definition of Composites

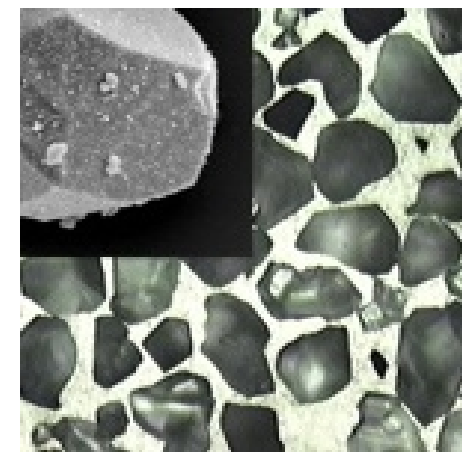
- § Composite materials are made of at least two distinct materials with different material parameters for each phase
- § Fiber reinforced composites (FRC) are composites where one material component (fiber) is used as a reinforcing material for the matrix
- § Particle reinforced composites consist of particles of one material dispersed in a matrix of a second material. Particles may have any shape or size, but are generally spherical, ellipsoidal, polyhedral, or irregular in shape



carbon fiber - epoxy



glass fiber - epoxy



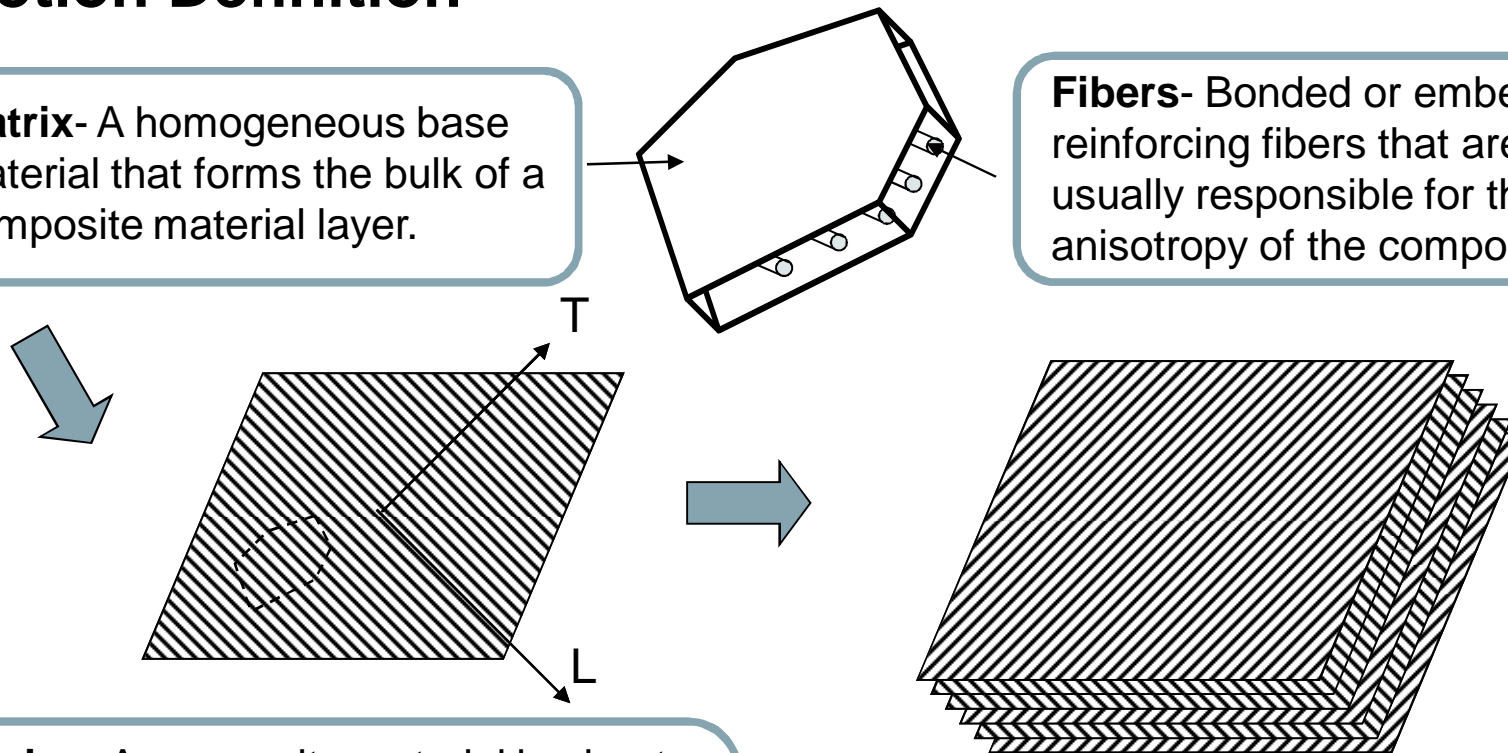
ceramic - aluminum

# Introduction to Composites

## Notion Definition

**Matrix**- A homogeneous base material that forms the bulk of a composite material layer.

**Fibers**- Bonded or embedded reinforcing fibers that are usually responsible for the anisotropy of the composite.



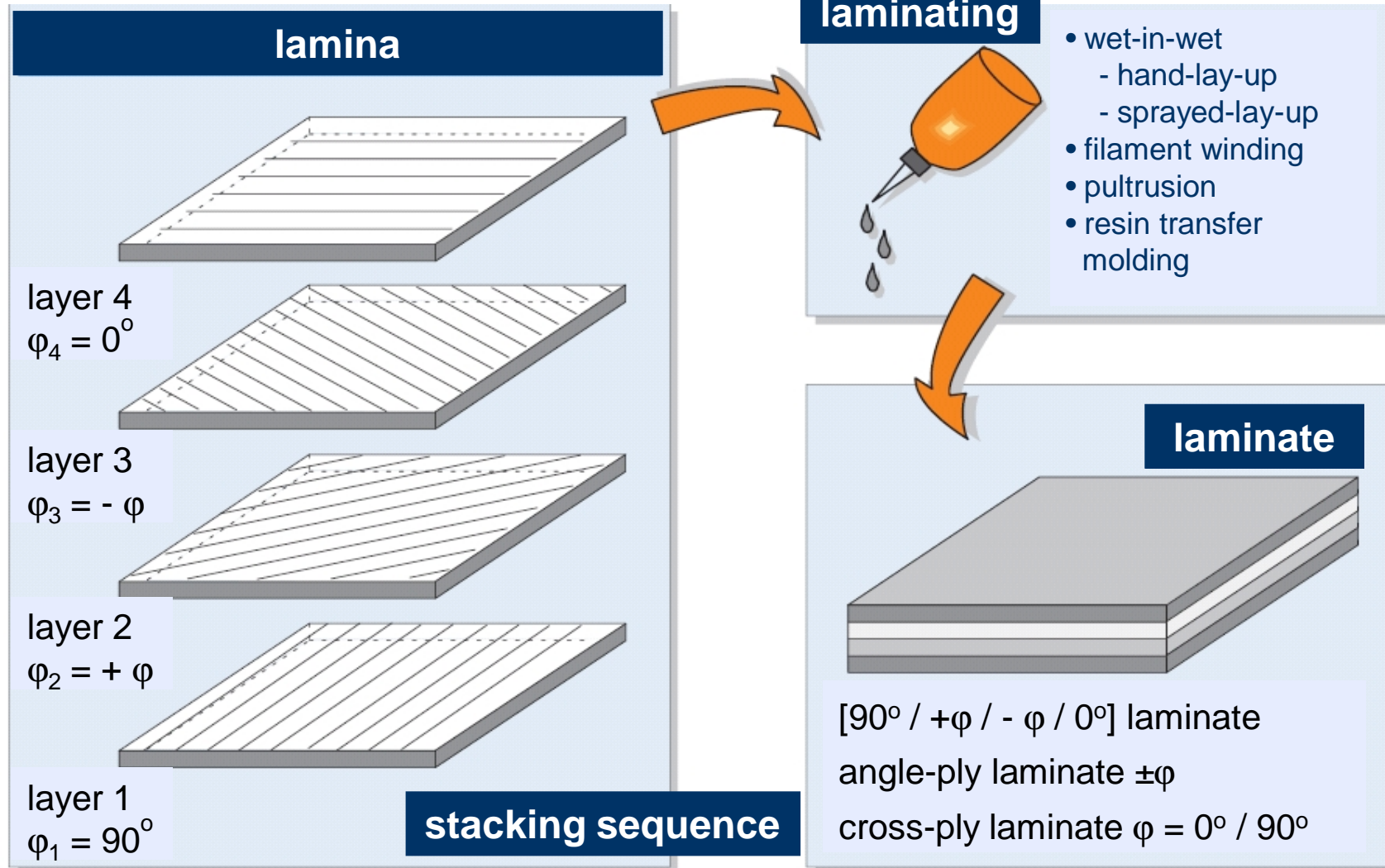
**Lamina**- A composite material in sheet form usually referred to as a **layer** or **ply**. The material properties of a layer is usually determined through an equivalent homogenization (smearing) process.

**Laminate**- A stack of lamina joined together in arbitrary directions, referred to as a composite **lay-up** or **stacking-sequence**.



# Introduction to Composites


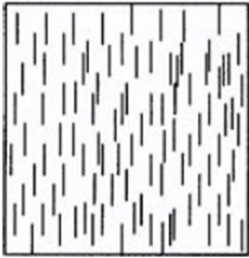

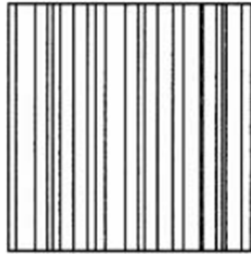
## Notion Definition



# Introduction to Composites

## Classification of Fiber Reinforced Composites

Raw Materials	Matrix	Fiber
	<ul style="list-style-type: none"><li>- Thermosets</li><li>- Thermoplastics</li><li>- Metals</li><li>- Ceramics</li></ul>	<ul style="list-style-type: none"><li>- Glass</li><li>- Carbon</li><li>- Aramid (Kevlar)</li><li>- Bor</li><li>- Ceramics</li></ul>

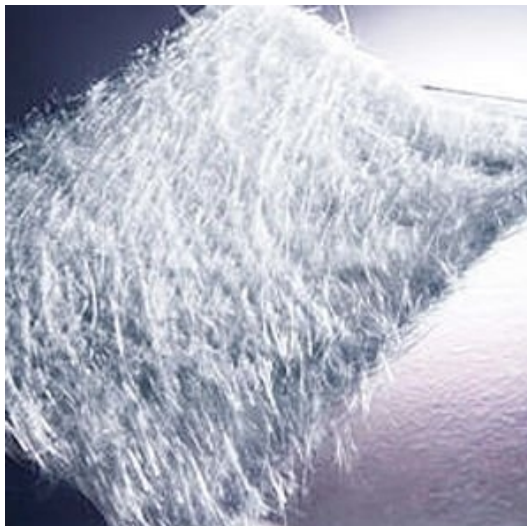
Types of Reinforcement	Short Fibers		Long Fibers	
	random	oriented	random	oriented
				

# Introduction to Composites

## Short Fiber/Particle vs. Endless Fiber Reinforced

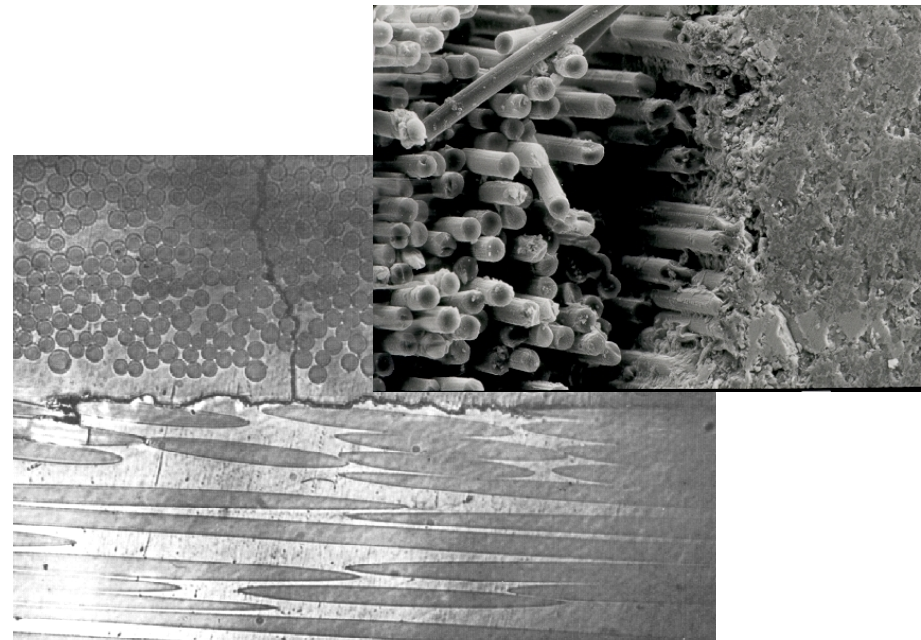
### § Particle and short fiber reinforced

- § Account for microstructure in form of size, shape, and orientation
- § Close gap in process chain for injection molding and mechanical simulation
- § Investigate local stresses and strains on fiber-matrix level



### § Long fiber reinforced, layered

- § Fast and efficient composite preprocessing
- § Composite optimized postprocessing with up-to-date failure criteria





# Introduction to Composites

## Failure Indicator and Reserve Factors

### § FPF – First-Ply-Failure Indicator

- § Mathematical equations indicate first failure of any ply
- § Failure modes predicted
- § Determines reserve factor, inverse reserve factor and margin of safety
- § Typical criteria: max stress, max strain, Tsai-Wu, Tsai-Hill, Hashin, Puck2D, Puck3D, Cuntze, LaRC



### § LPF – Last-Ply-Failure

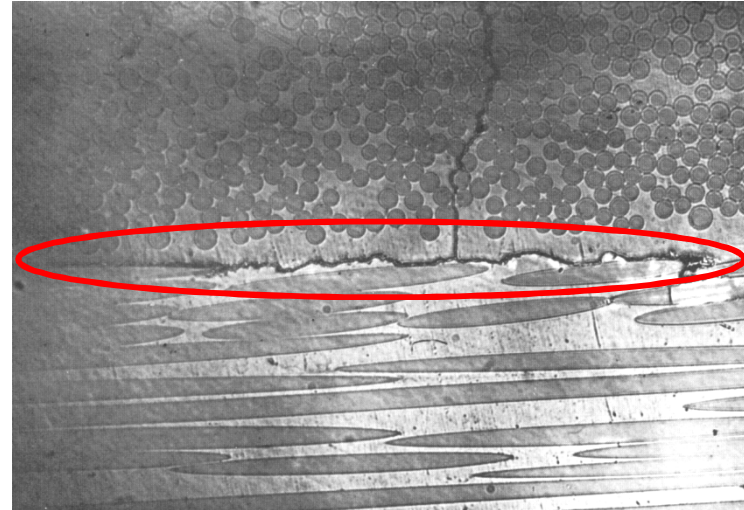
- § Further loading beyond FPF until ultimate failure of laminate
- § Post-failure formulations needed (degradation, ply-discount method)
- § May also include energy dissipating methods

# Introduction to Composites

## Special Failure Modes in Layered Composites

### § Delamination

- § Failure of interface between two plies in normal direction under tension
  - à interlaminar failure
- § Normal stress in thickness direction is driver
  - à special method in shells needed

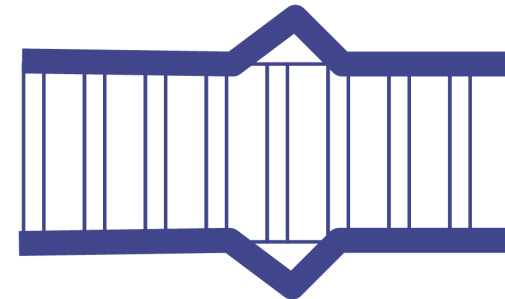


# Introduction to Composites

## Special Failure Modes in Sandwich Structures

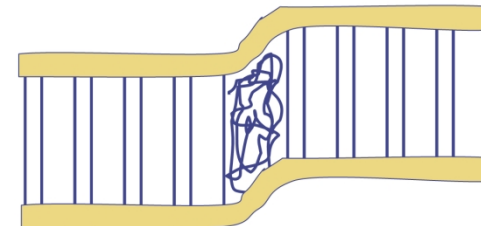
### § Wrinkling

- § local buckling of a face sheet under compression
- § failure indicator also possible with shell type of modeling for whole sandwich



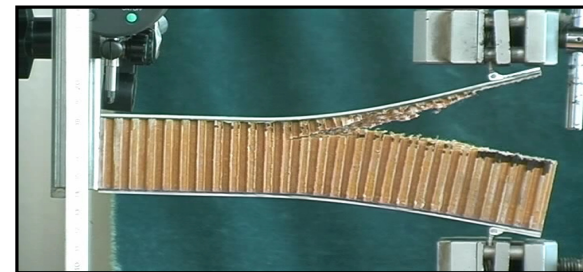
### § Core failure

- § local failure of core in shear or tensile loading
- § failure indicator also possible with shell type of modeling for whole sandwich



### § Debonding

- § Failure in interface between face sheet and core of sandwich structures
- § Only predictable when core and sheet are separately modeled





$$\bar{\Pi} = \frac{1}{2} \sum_e \{u\}^T \cdot [K] \cdot \{u\} - \{u\}^T \cdot \{F\}$$

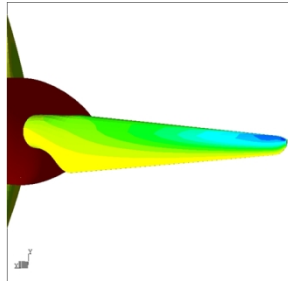
FEM SOFTWARE AND SERVICES



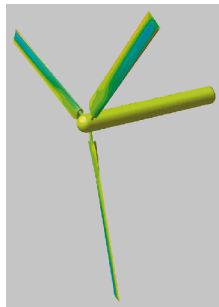
# ANSYS in Wind Power Industry

# ANSYS in Wind Power Industry

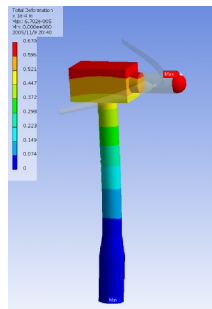
## Multiple Physics & Coupling



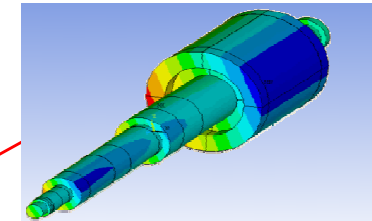
Blade design and performance



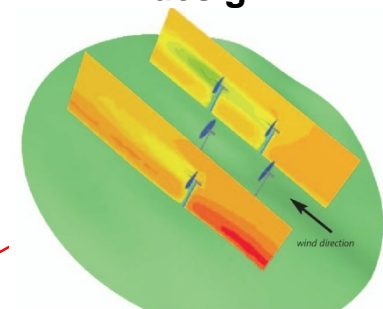
Rotor sizing and acoustics



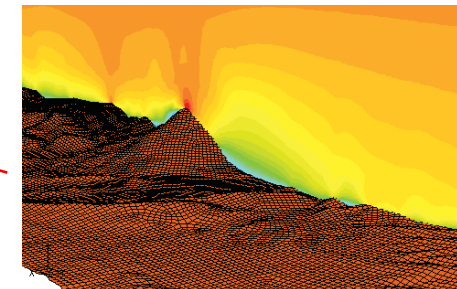
Tower design and FSI



Generator and shaft design



Wind farm configuration for optimal power generation

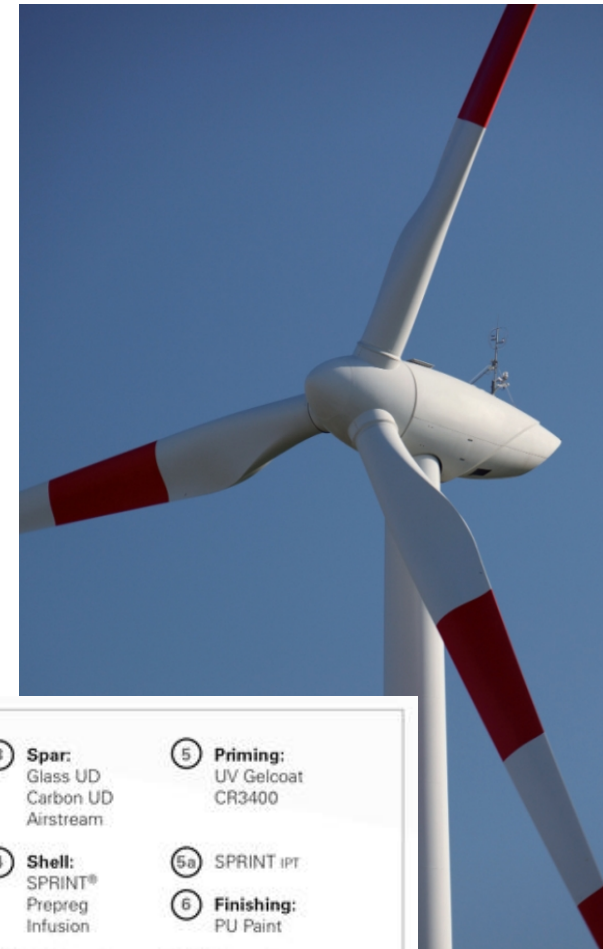
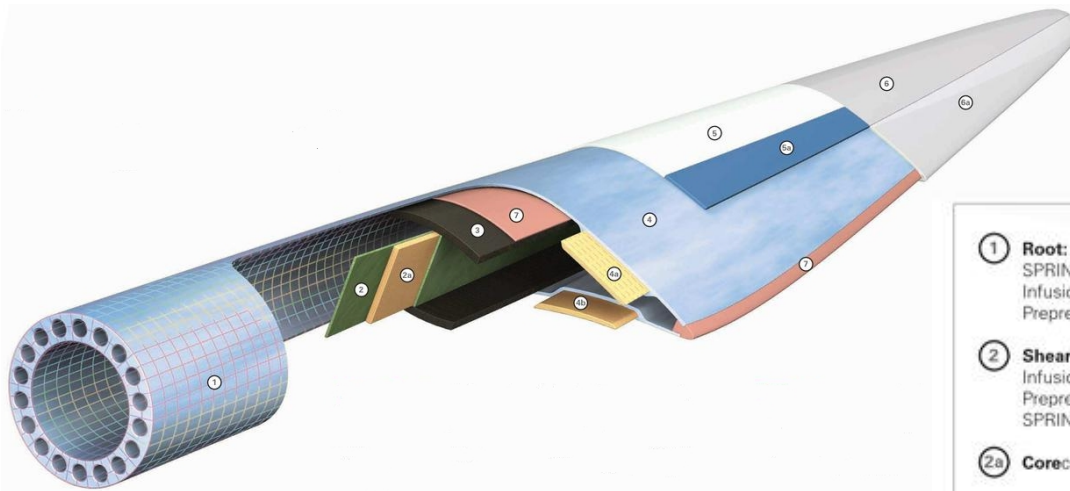


Site selection, land and sea

# ANSYS in Rotor Blade Design

## Challenges in Structural Modeling

- § Laminate & Sandwich
- § Free form surface from aerodynamics
- § Multiple material systems involved
- § Complex layup with huge amount of plies
- § Arbitrary fiber orientation
- § Ply and sandwich failure



- |  |  |   |
|--|--|---|
| ① <b>Root:</b><br>SPRINT®<br>Infusion<br>Prepreg     | ③ <b>Spar:</b><br>Glass UD<br>Carbon UD<br>Airstream | ⑤ <b>Priming:</b><br>UV Gelcoat<br>CR3400         |
| ② <b>ShearWeb:</b><br>Infusion<br>Prepreg<br>SPRINT® | ④ <b>Shell:</b><br>SPRINT®<br>Prepreg<br>Infusion    | ⑤a SPRINT IPT                                     |
| ②a <b>Corecell™</b>                                  | ④a Infusion Core                                     | ⑥ <b>Finishing:</b><br>PU Paint                   |
|  | ④b Prepreg Core                                      | ⑥a Epoxy Gelcoat                                  |
|  |  | ⑦ <b>Structural Adhesive:</b><br>SP340<br>SP340LV |

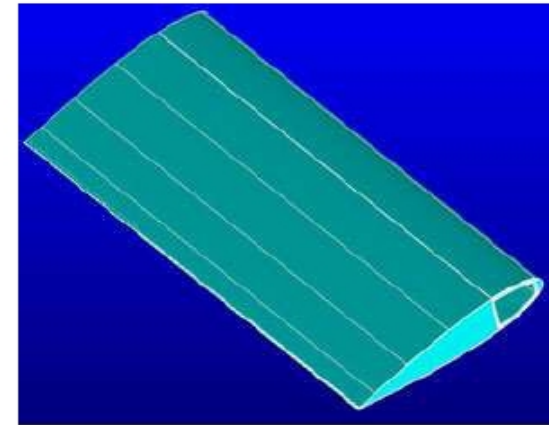


# ANSYS in Rotor Blade Design

## Levels of Structural Modeling

### § Beam Models

- § Fast investigation in preliminary design phase
- § Using variational-asymptotic method to reduced model order



**CADFEM**

# ANSYS in Rotor Blade Design

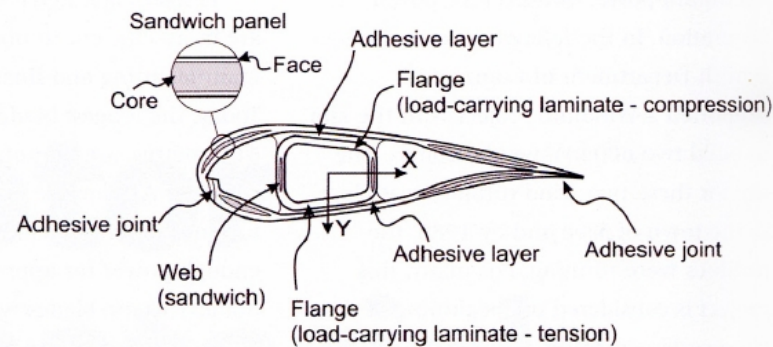
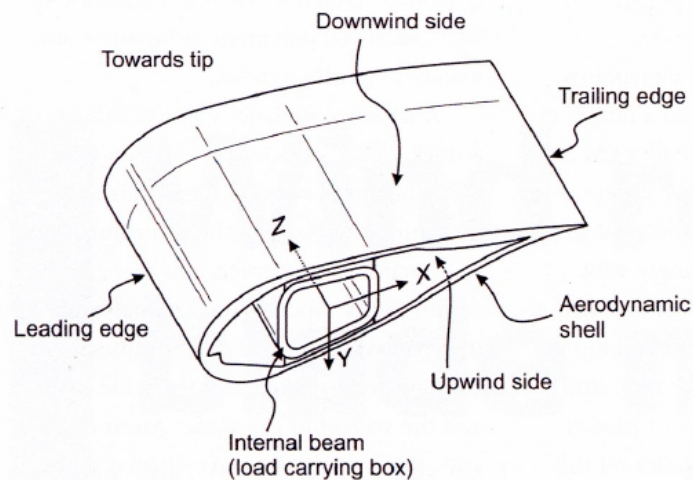
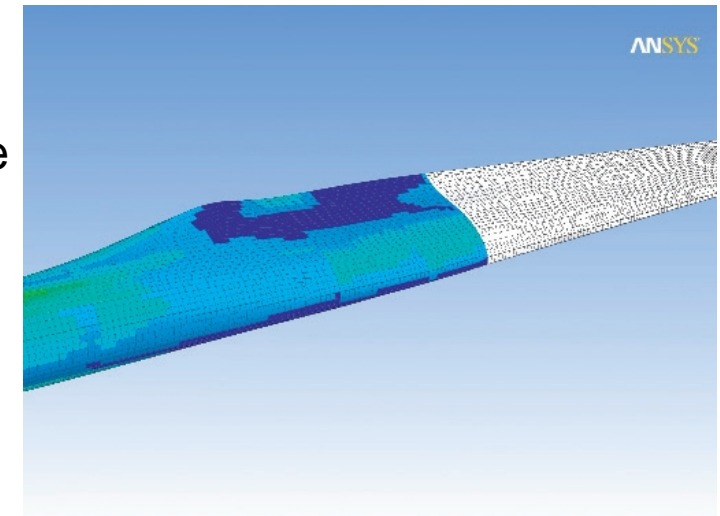
## Levels of Structural Modeling

### § Beam Models

- § Fast investigation in preliminary design phase
- § Using variational-asymptotic method to reduced model order

### § Shell Models

- § Determine laminate layout in stiffness & strength evaluations
- § Nonlinear simulations including buckling



# ANSYS in Rotor Blade Design

## Levels of Structural Modeling

### § Beam Models

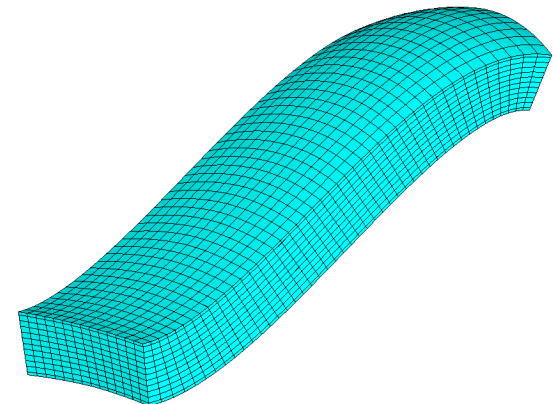
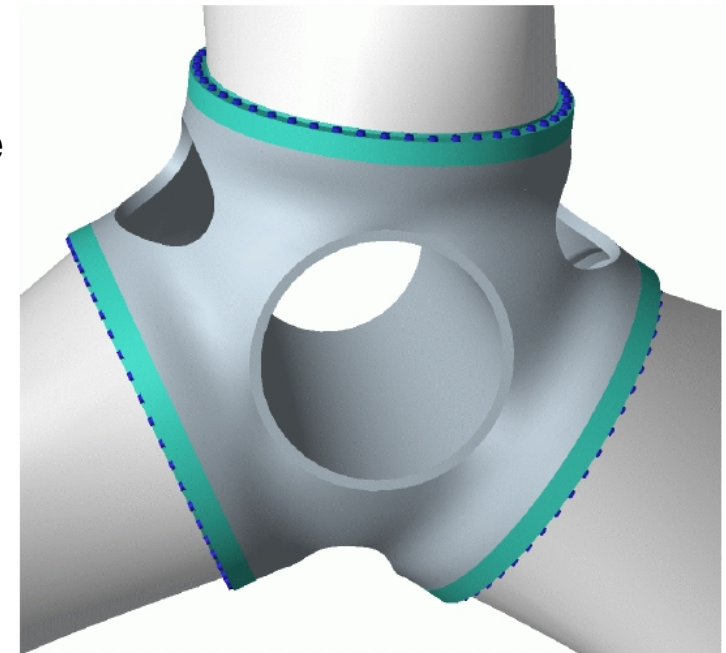
- § Fast investigation in preliminary design phase
- § Using variational-asymptotic method to reduced model order

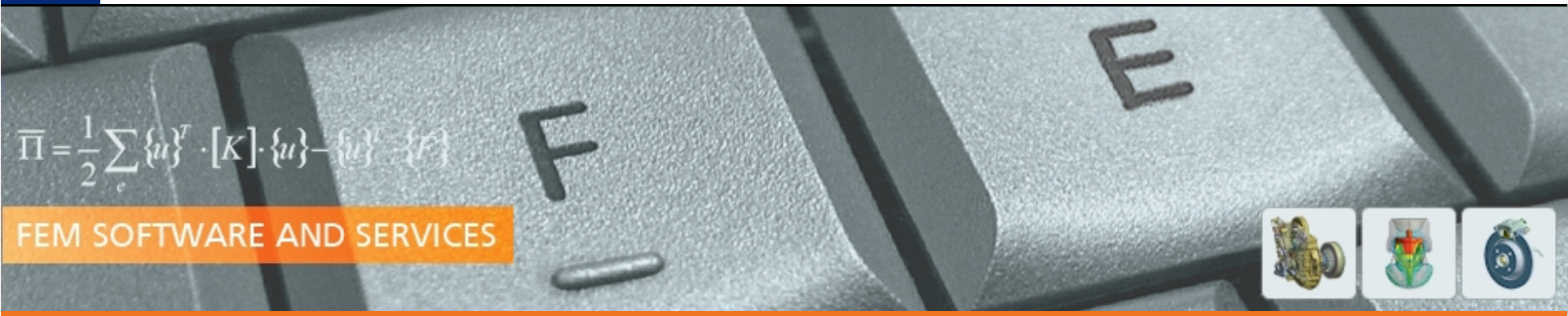
### § Shell Models

- § Determine laminate layout in stiffness & strength evaluations
- § Nonlinear simulations including buckling

### § Solid Models

- § Detailed investigation of load application areas, supports, inserts
- § Investigation of 3D stress state in curved regions and thick laminates




$$\bar{\Pi} = \frac{1}{2} \sum_e \{u\}^T \cdot [K] \cdot \{u\} - \{u\}^T \cdot \{F\}$$

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# Solutions for Layered Composites

Efficient and Closed Design Loop

**ANSYS**<sup>®</sup>

ANSYS Competence Center FEM

**CADFEM**<sup>®</sup>

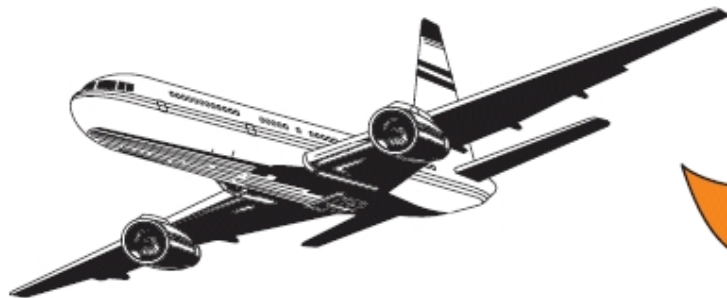


# Levels of Observation

including material models for first-ply-failure and delamination

**ANSYS**

goal: FEM—simulation of shell-like structures

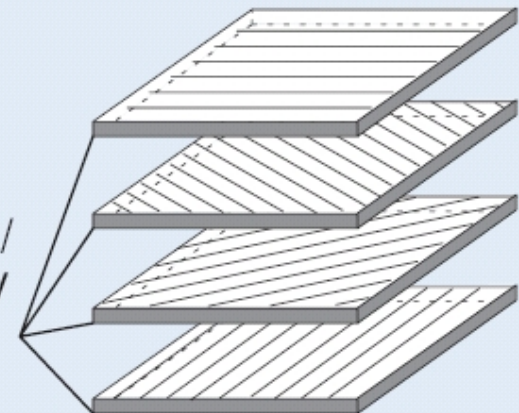


**structural-level**

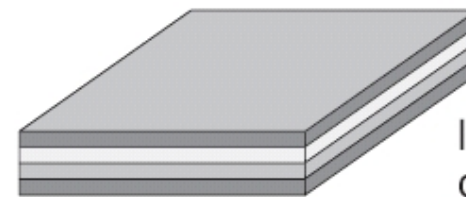
**layer-level**

homogenised layers

orthotropic / transversely isotropic material behaviour

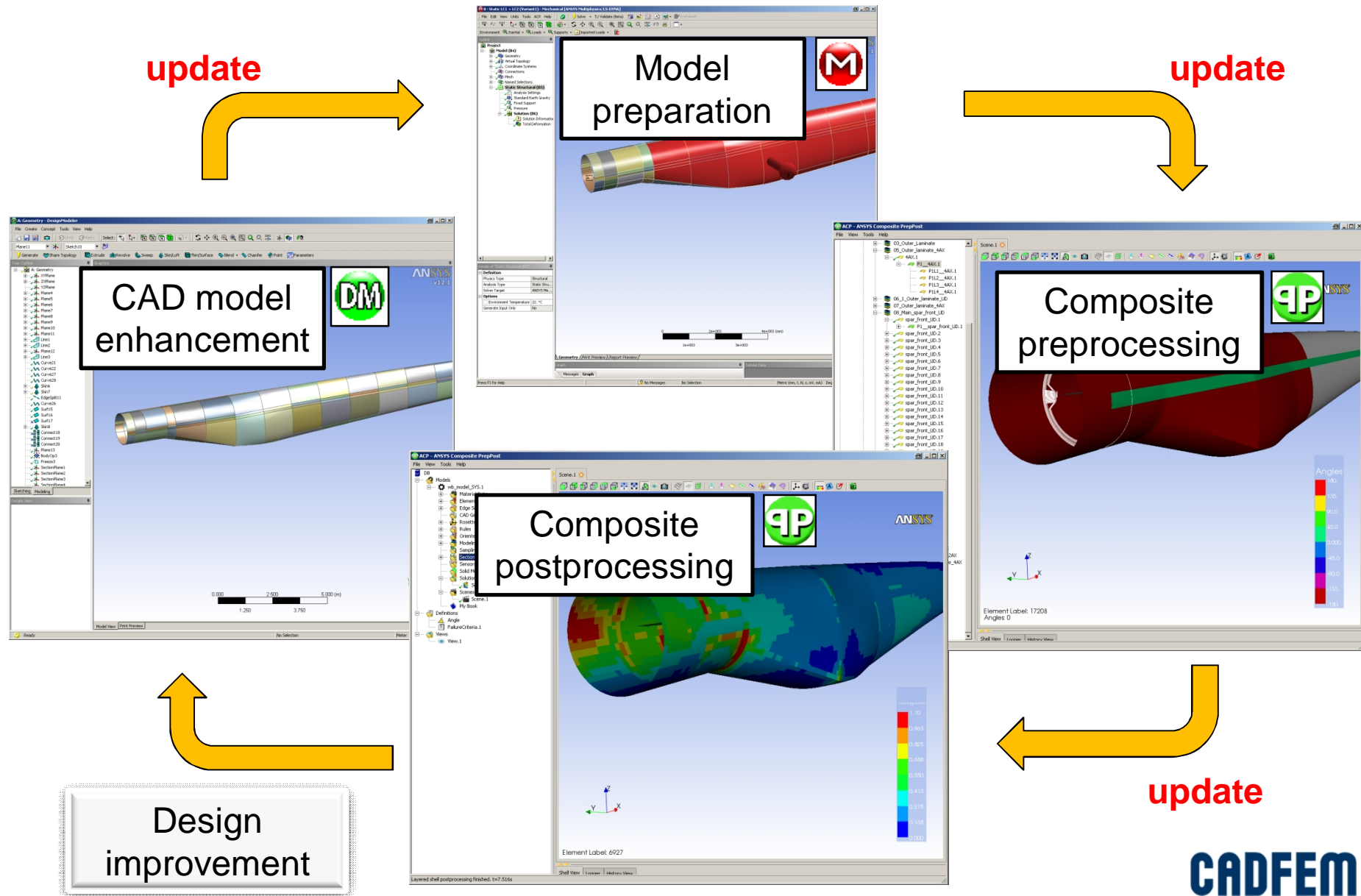


laminated cross section



**laminate-level**

# Simulation-Driven-Product-Design



# ANSYS FE Solver Features

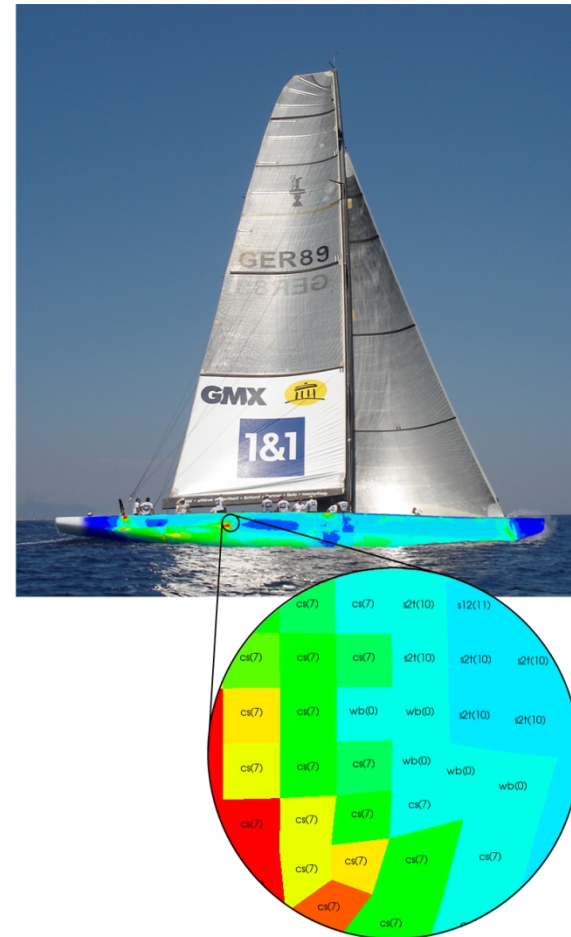
## ANSYS Solver Features

- § Transversely isotropic and orthotropic material models
  - § Layered shell, solid and beam elements
  - § Tapering cross section for shells and beams
  - § Arbitrary shell offset (reference surface)
  - § Sandwich option for shells to account for soft core
  - § Some basic FPF criteria (max stress, max strain, Tsai-Wu)
  - § Delamination & Debonding with energy release using interface elements
- 
- à Powerful solver features available
  - à Efficient pre- and postprocessing suitable for layered composites through ANSYS Composite PrepPost

# ANSYS Composite PrepPost

## Introduction to ANSYS Composite PrepPost

- § Dedicated to allow for easy and efficient definition and modification of **layered-composite** structures as well as their specific post-processing
- § Integrated in ANSYS Workbench as well as in ANSYS Classic
- § Dedicated to provide leading-edge technology in pre- and postprocessing of layered composite structures

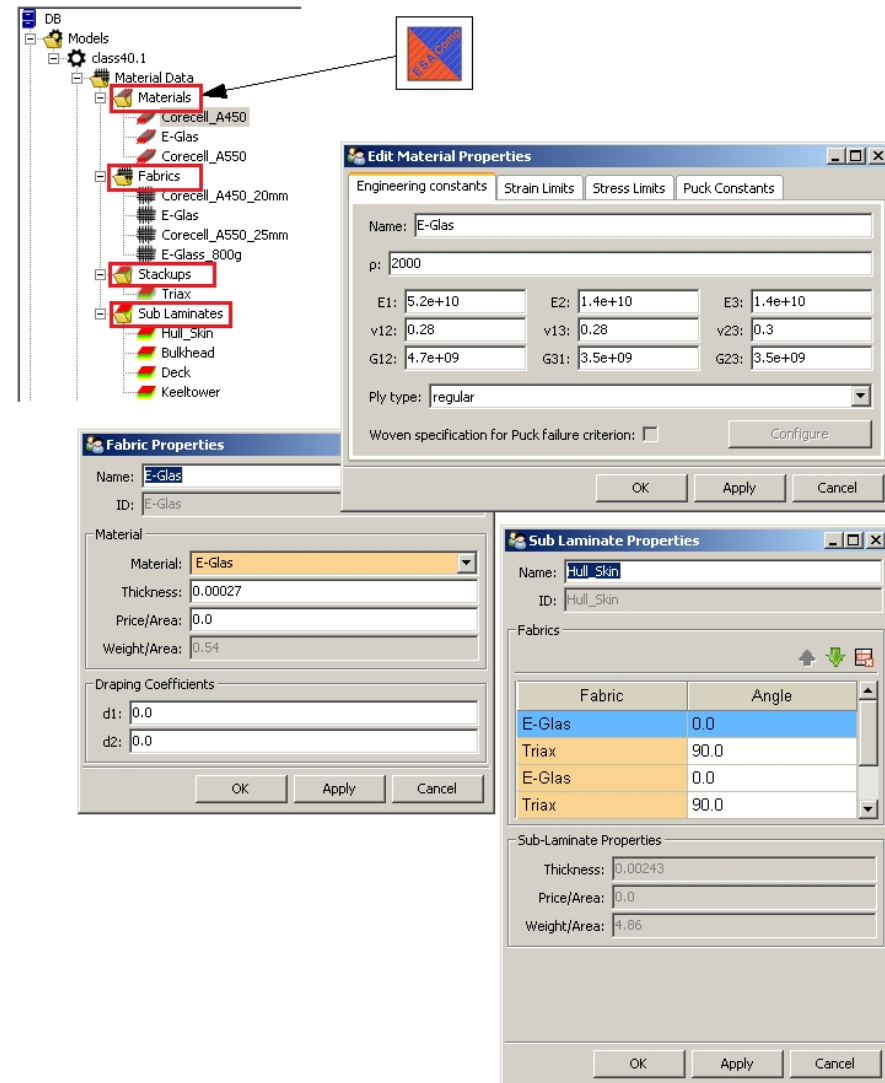




# ANSYS Composite PrepPost

## Key Features for Composites Preprocessing

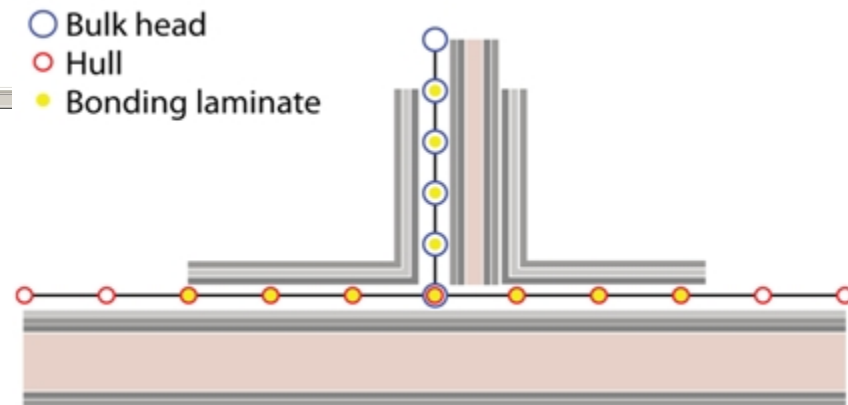
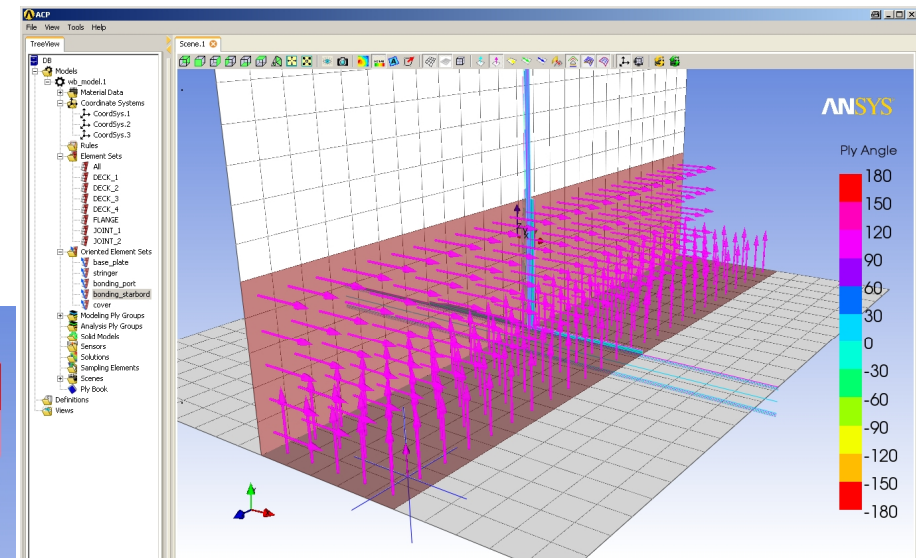
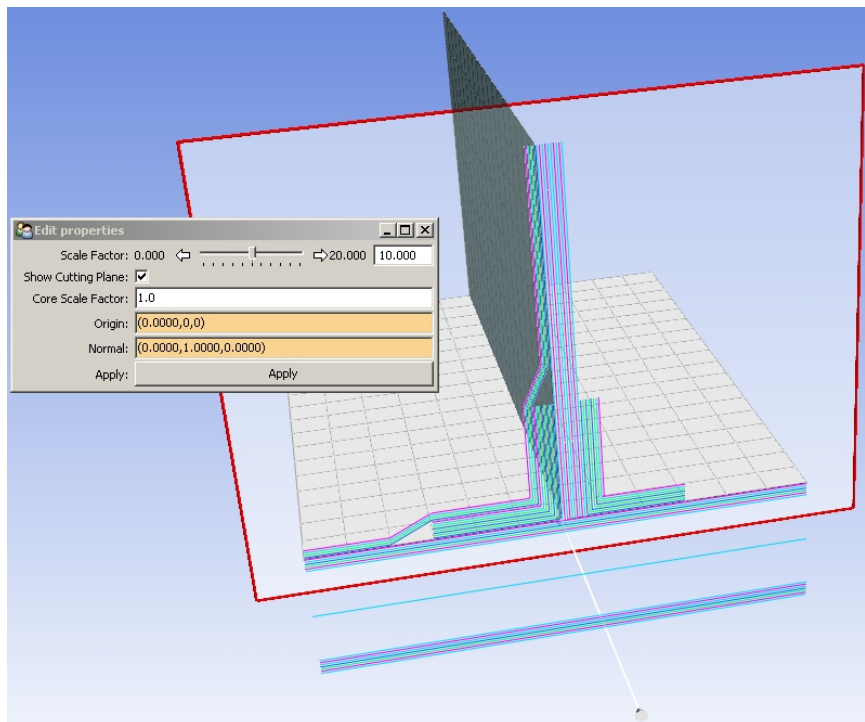
- § Unique material definition functionality tailored for modeling layered composite structures
- § Fabrics, Stack-ups and Sub-laminates as modeling plies
- § Basis engineering data and additional advanced post-processing data provided
- § Import material data base from ESAComp



# ANSYS Composite PrepPost

## Key Features for Composites Preprocessing

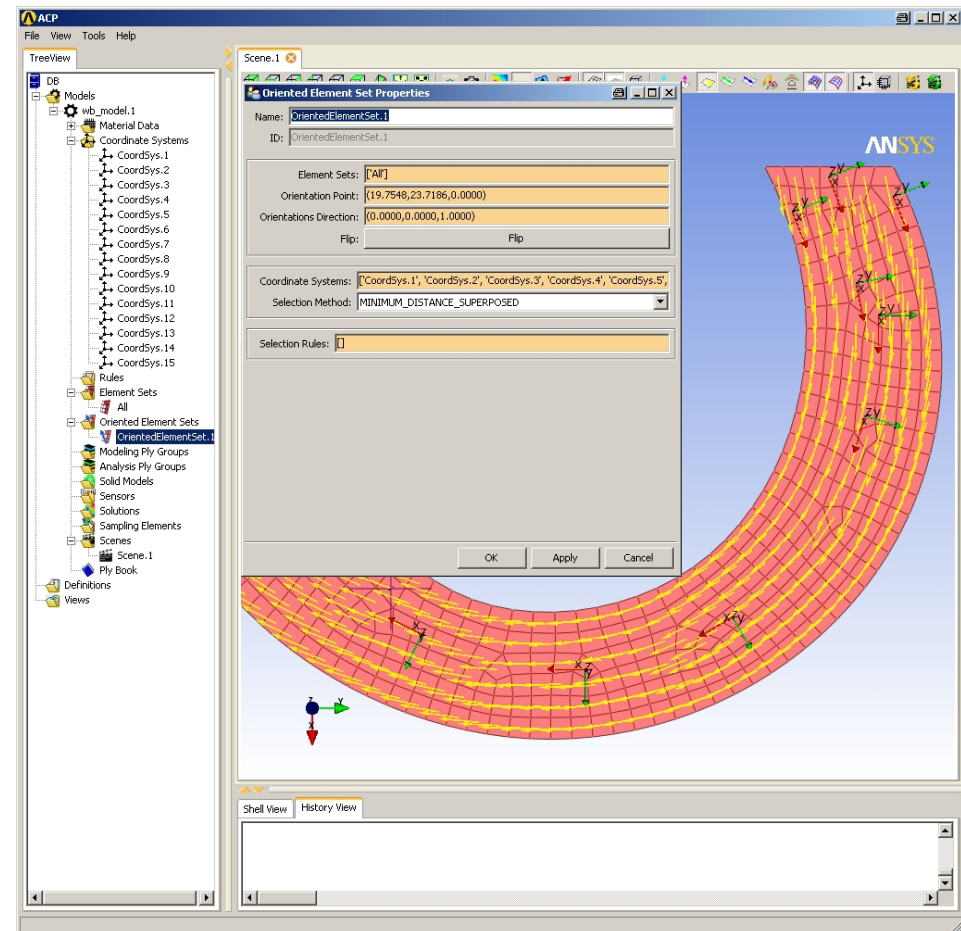
- § Arbitrary material application direction and material 0° direction
- § Multiple material application direction for easy asymmetric laminate definition



# ANSYS Composite PrepPost

## Key Features for Composites Preprocessing

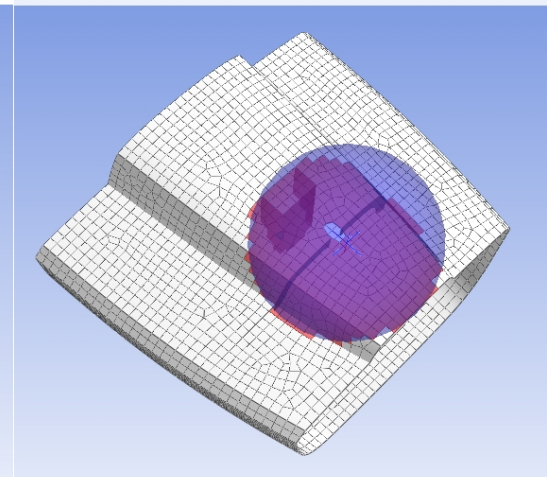
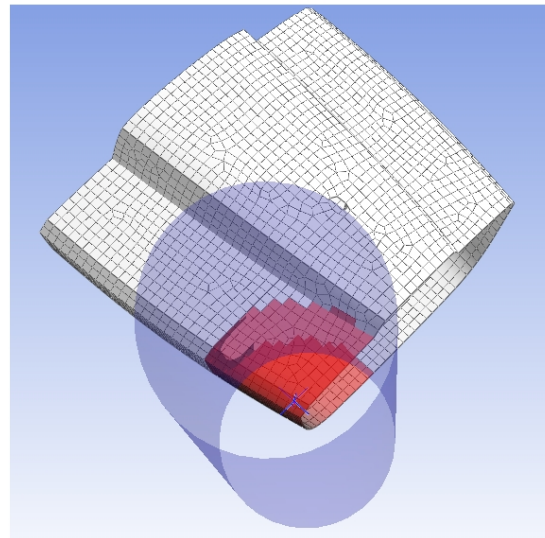
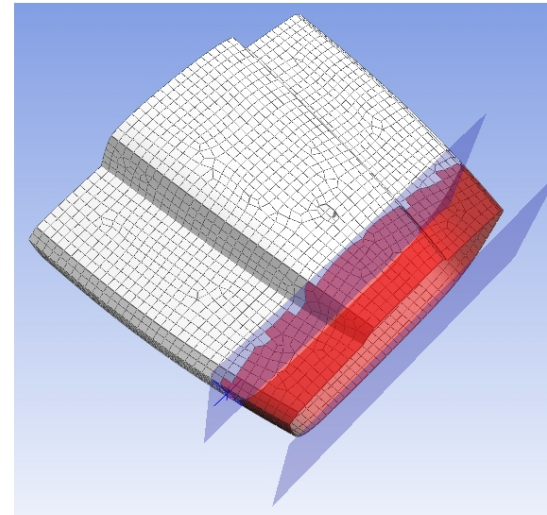
- § Cartesian, cylindrical and spherical coordinate systems for material reference direction
- § Arbitrary material reference direction through multiple coordinate systems and interpolation



# ANSYS Composite PrepPost

## Key Features for Composites Preprocessing

§ Selection rules provide practical means to reselect group of elements for which the physical plies are applied

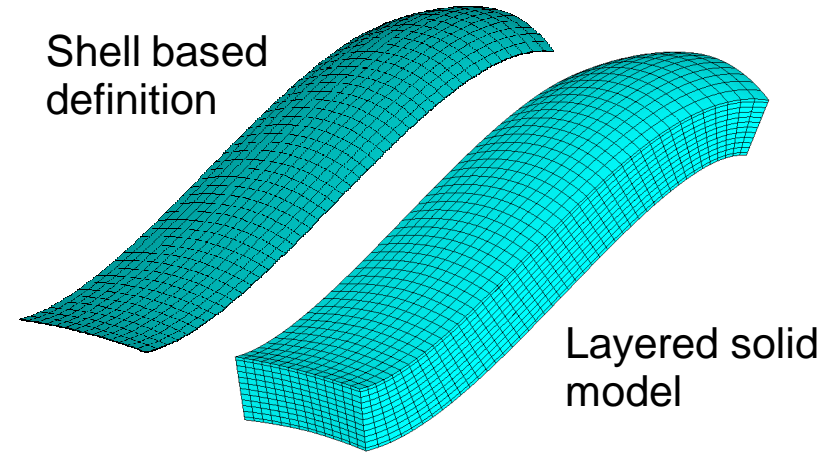




# ANSYS Composite PrepPost

## Key Features for Composites Preprocessing

- § Generation of layered-solid composite models in same way as layered shell-models
- § Layer-wise post-processing of layered-solid models identically to the post-processing of layered shell-models

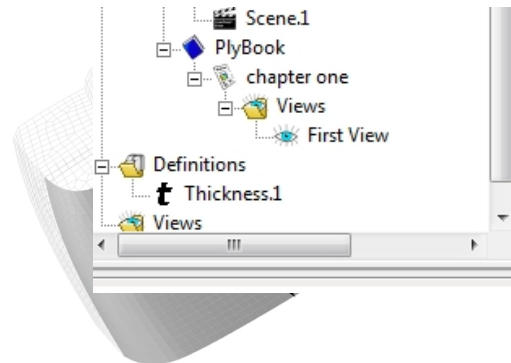


# ANSYS Composite PrepPost

## Key Features for Composites Preprocessing

- § Draping functionality to predict ability to drape and fiber-angle correction
- § Flat-wrap of plies in \*.dxf and boundaries as \*.stp & \*.iges
- § Generation of individually formatted ply-books and export in various formats (\*.html, \*.pdf, \*.odt...)

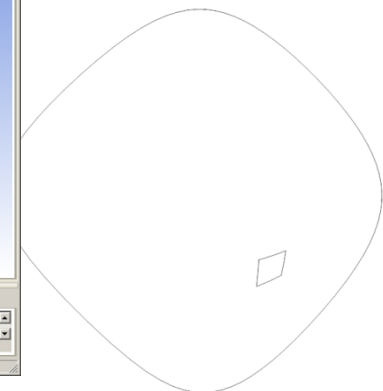
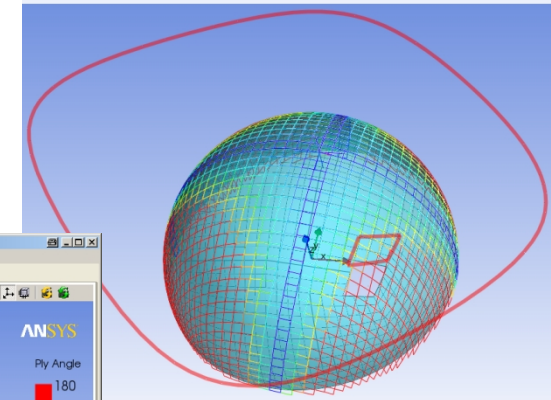
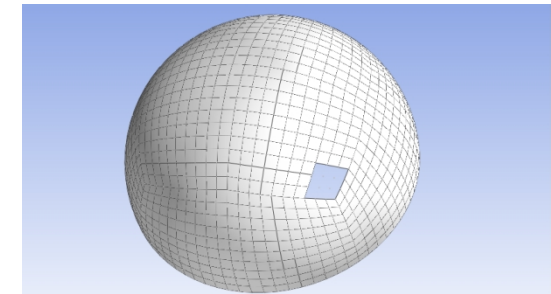
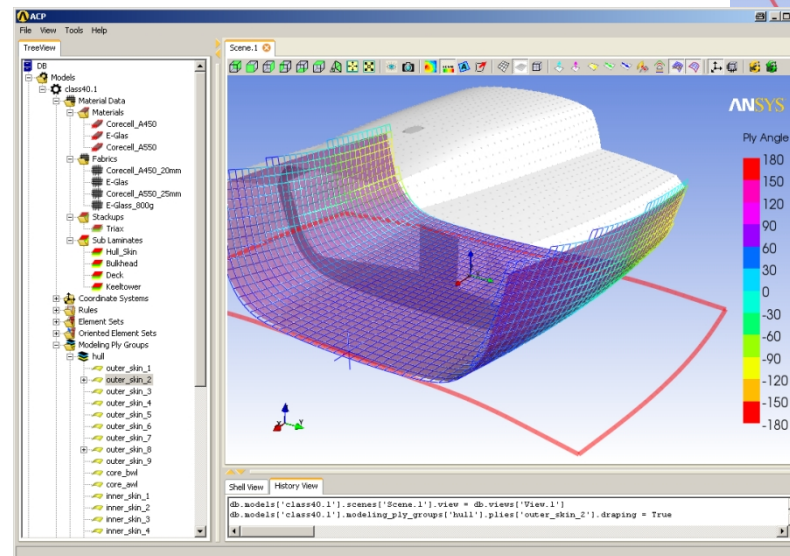
Ply 12/51



Picked element ID: None, Scalar: None

Thu Jan 15 09:03:11 2009

Ply Name:	Bottom
Ply ID:	Bottom.3
Ply Nr:	12/51
Material:	UD_300/35
Orientation:	45
Thickness:	0.3
Ply Area:	20
Ply Weight:	5721
Ply Cost:	28.22



# ANSYS Composite PrepPost

## Key Features for Composites Postprocessing

§ Evaluation of Inverse Reserve Factors (IRF's) for several composite failure criteria at all integration points of all layers of all requested layered Finite Elements.

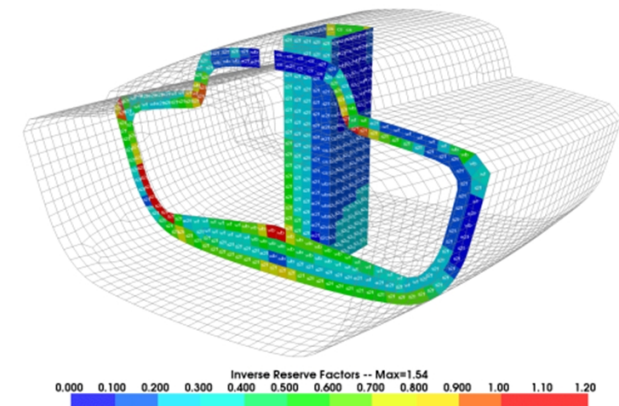
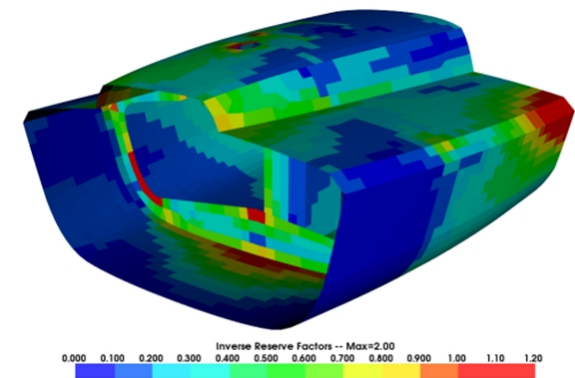
§ Definition and configuration of arbitrary combinations of failure criteria.

Currently available:

- § Tsai-Wu, Tsai-Hill, Hashin
- § Puck 2D and 3D for UD and weave materials
- § LaRC (Langley Research Center)
- § Core failure and face sheet wrinkling for sandwich structures

§ ACP computes four result values per in-plane data point:

- § Maximum IRF of all criteria of all layers
- § Active failure mode (criterion with highest IRF)
- § Layer index where highest IRF occurs
- § Critical load case



# ANSYS Composite PrepPost

## Key Features for Composites Postprocessing

§ Text plot highlights critical failure mode, layers and load case

§ Failure modes

§  $tw$  = Tsai-Wu

§  $s1t$  = Fiber failure due to tension

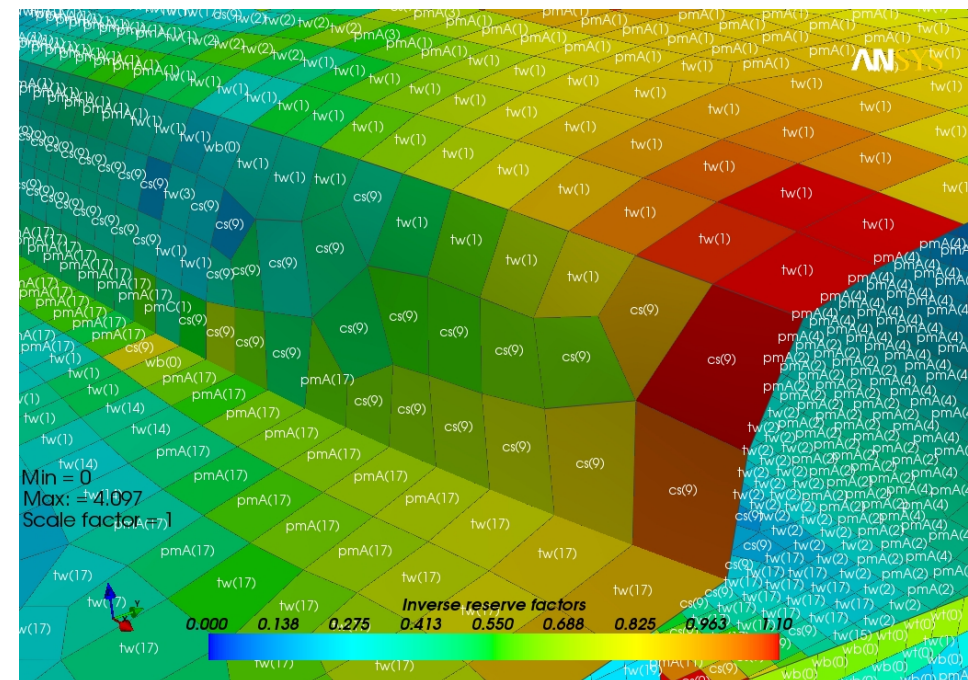
§  $s2c$  = Matrix failure due to compression

§  $cs$  = core failure

§ Critical layer

§ (7) = layer 7

§ Critical load case

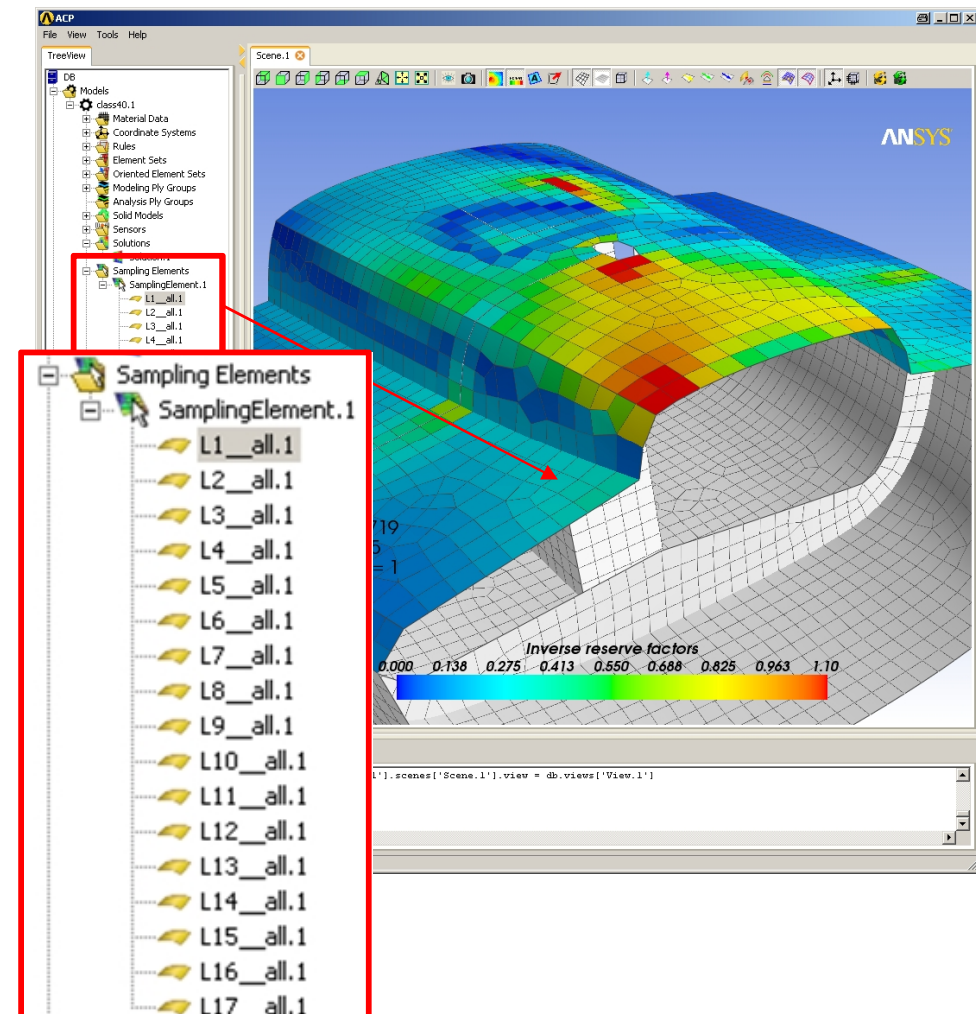
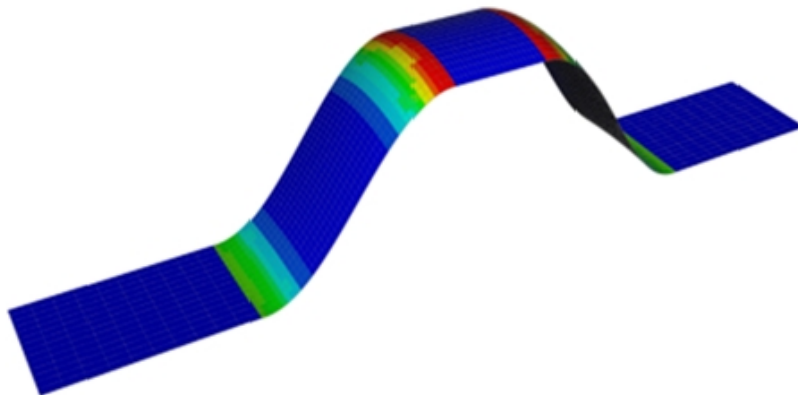




# ANSYS Composite PrepPost

## Key Features for Composites Postprocessing

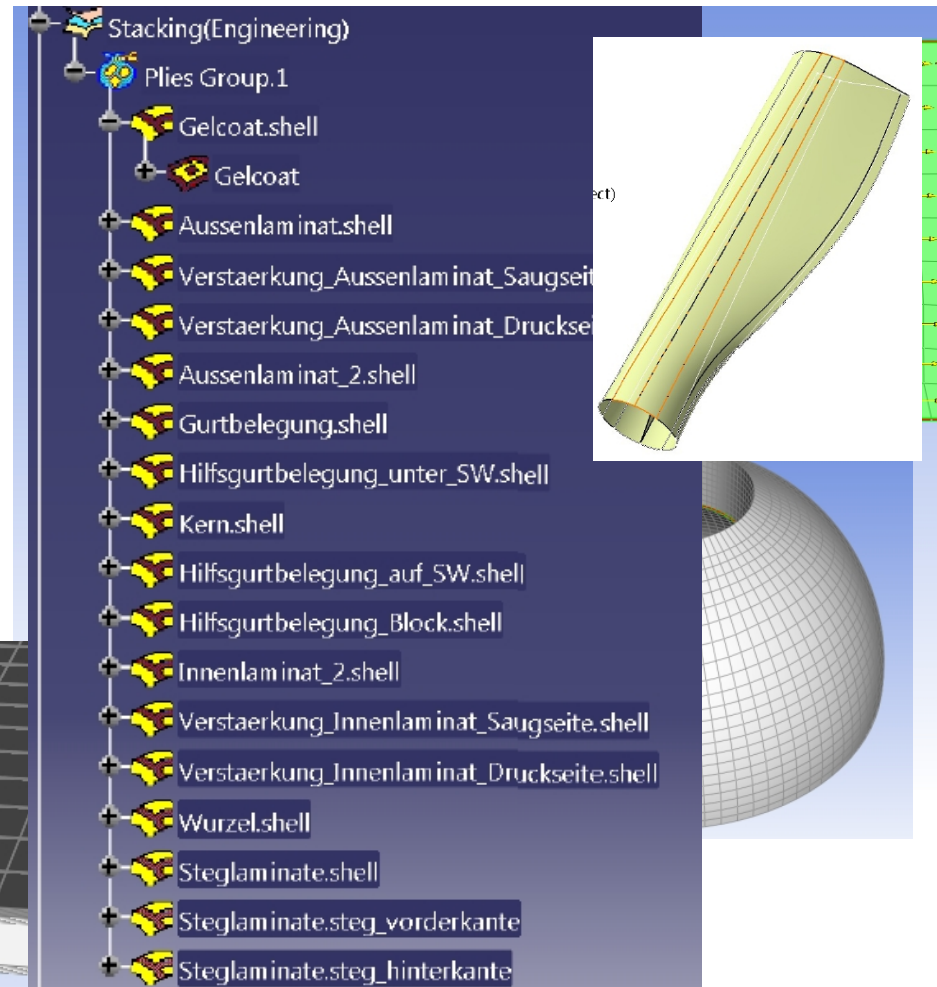
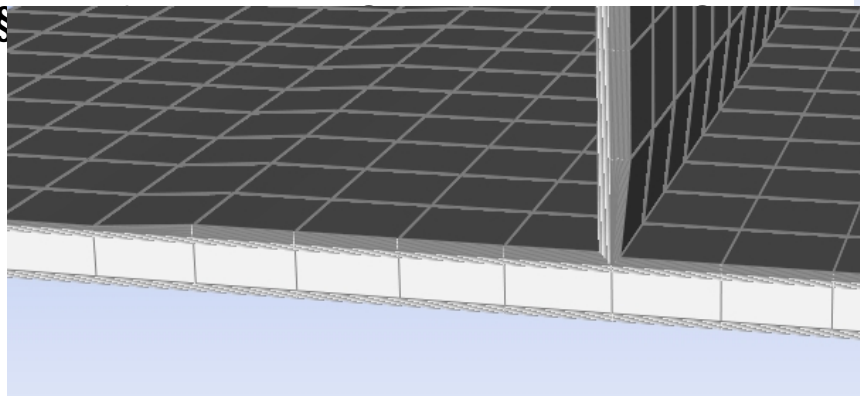
- § Sampling element provides stacking sequence (laminate) of specific element
- § Enables ply-based strain and stress visualization.
- § Each ply of laminate can be select from ply-list for result visualization
- § Normal thickness stress calculation in shell elements



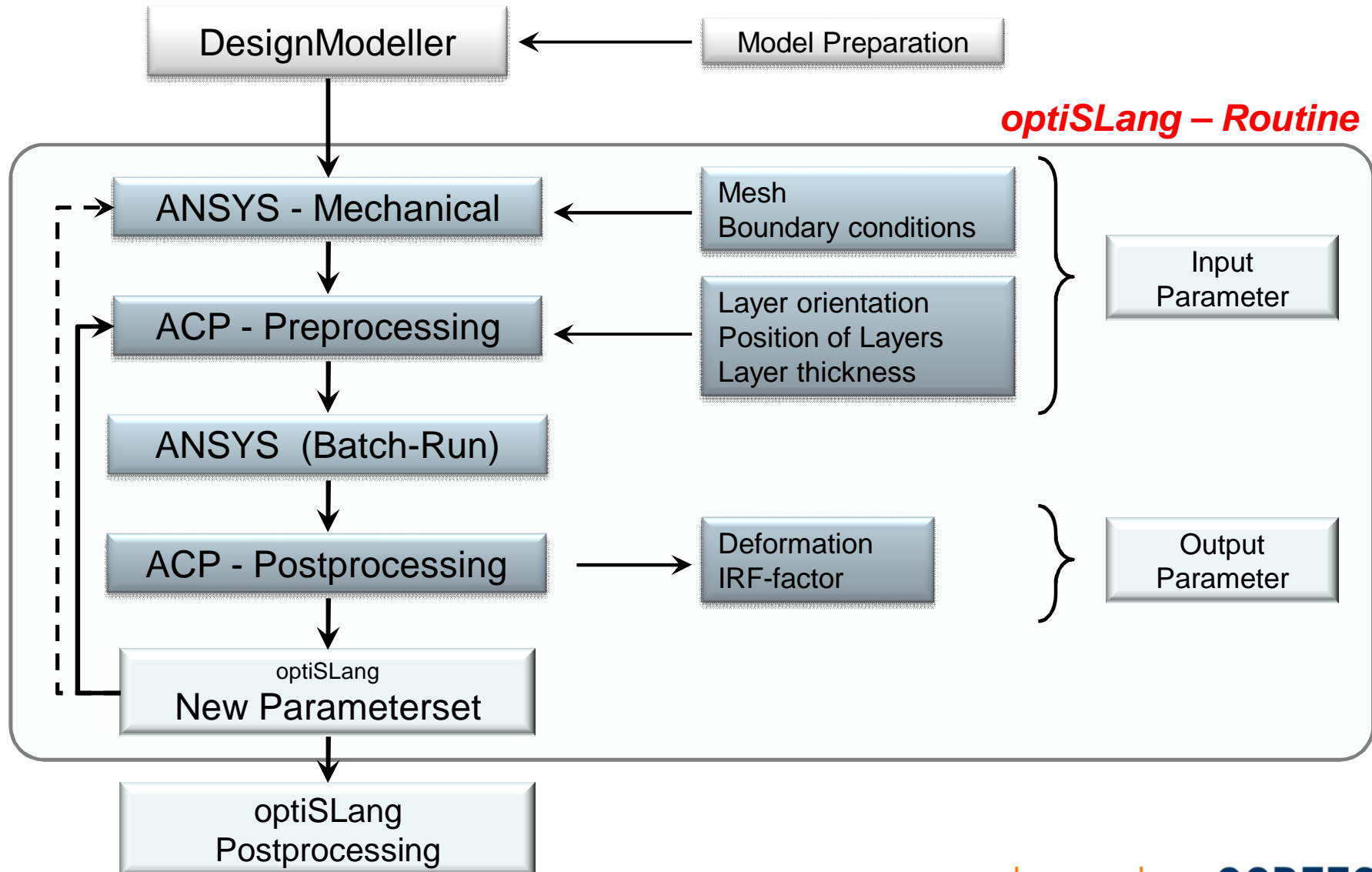
# Latest Features

## Latest Features in ANSYS Composite PrepPost

- § Shaped cores using CAD-surfaces for variable core thickness
- § Concept of modeling, production and analysis plies
- § Multiple section cuts with full control on visualization of modeling, production and analysis plies
- § Tube rules on CAD-lines for element selection
- § CAD-lines for material reference direction
- § Extended solid extrusion features

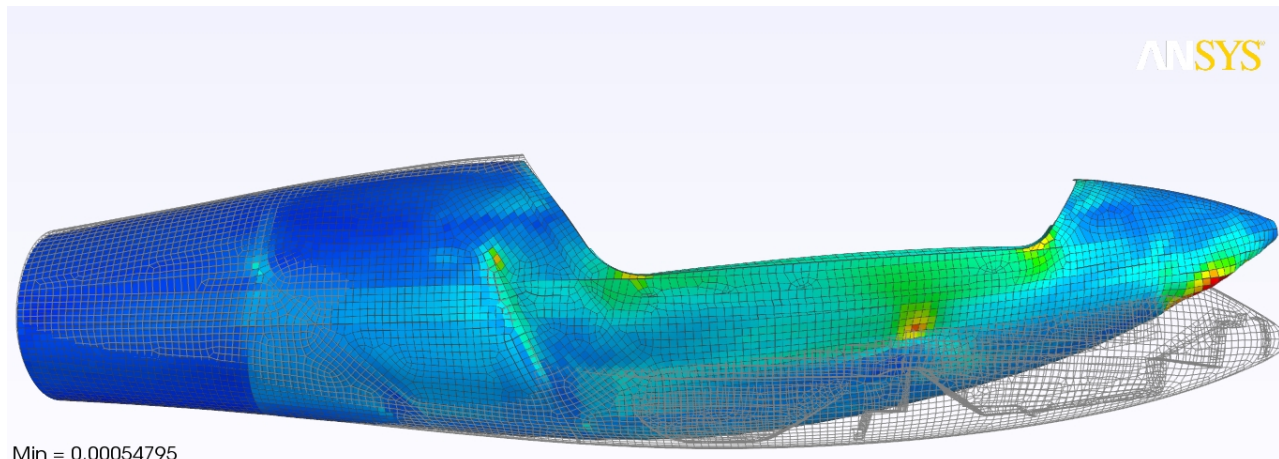


# Composites & Optimization



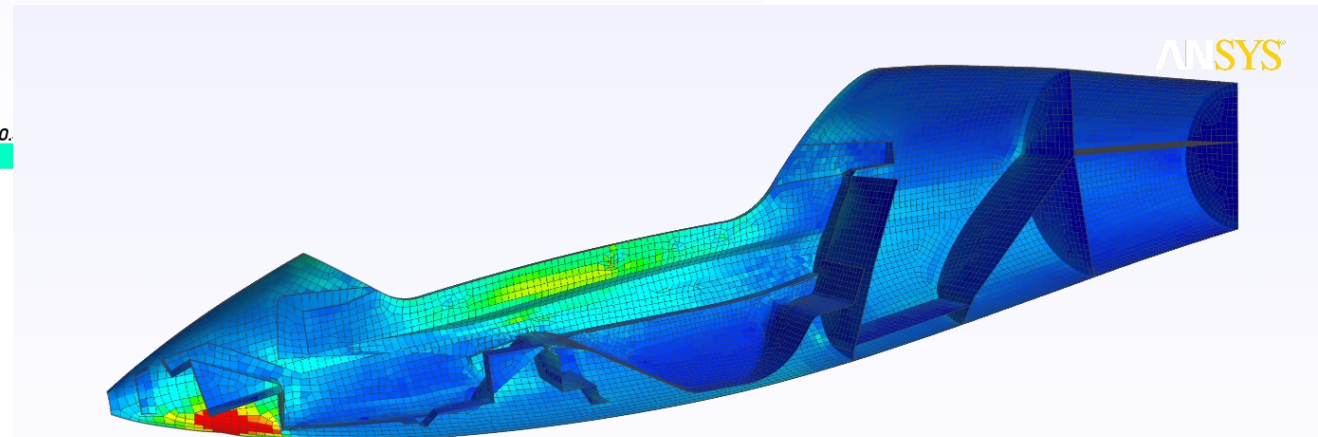
# Composites & Optimization

## Cockpit of Sail Plane



Min = 0.00054795  
Max: = 4.8566  
Scale factor = 10

0.000 0.138 0.275 0.413

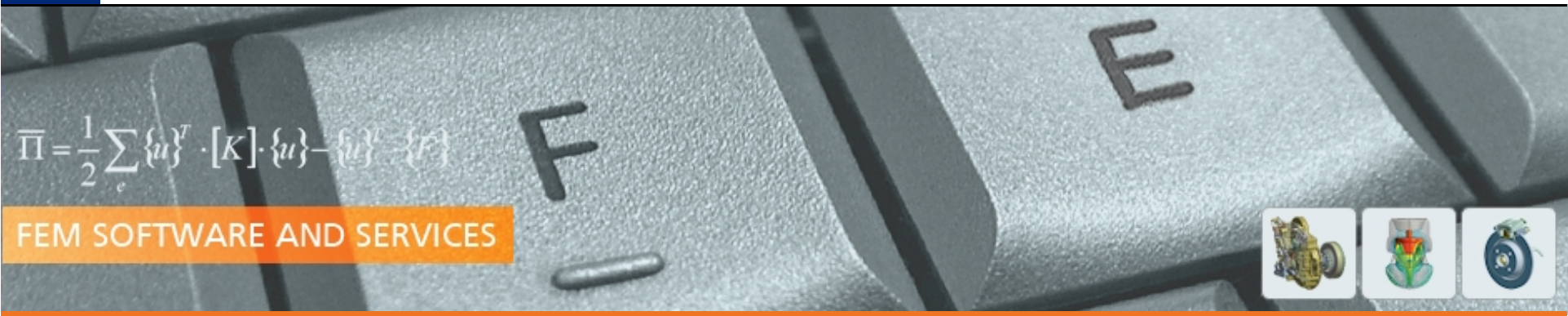


Min = 0.00054795  
Max: = 4.8566  
Scale factor = 1

0.000 0.138 0.275 0.413 0.550 0.688 0.825 0.963 1.10

Inverse reserve factors




$$\bar{\Pi} = \frac{1}{2} \sum_e \{u\}^T \cdot [K] \cdot \{u\} - \{u\}^T \cdot \{F\}$$

FEM SOFTWARE AND SERVICES



# Live-Demonstration

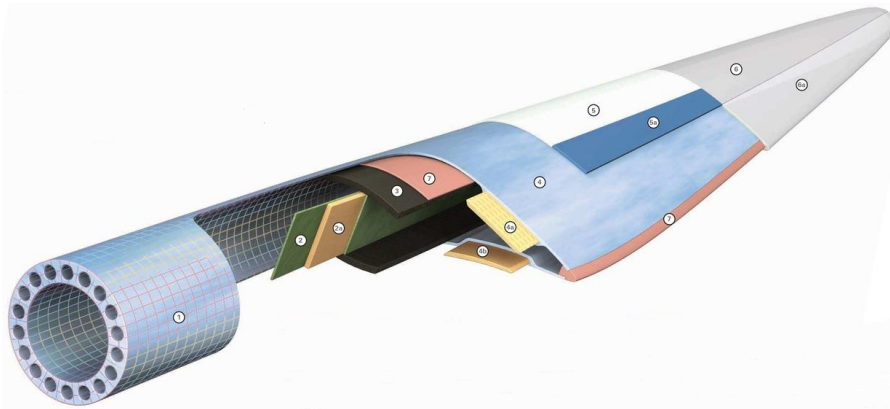
Rotorblade

# Summary

## Composites – Catch the Wind!

### § Particle and short fiber reinforced

- § Account for microstructure in form of size, shape, and orientation
- § Close gap in process chain for injection molding and mechanical simulation
- § Investigate local stresses and strains on fiber-matrix level
- § **Solution: ANSYS + Digimat**



### § Long fiber reinforced, layered

- § Fast and efficient composite preprocessing
- § Composite optimized postprocessing with up-to-date failure criteria
- § **Solution: ANSYS + ANSYS Composite PrepPost**

